

RL-23 Combustor Test Facility

The RL-23 combustor test facility (Fig. 1) is located in the Rocket Lab area of the center. This high temperature/low pressure facility is currently used in support of the NASA High Speed Research (HSR) Program focusing on supersonic test conditions, and the NASA Advanced Subsonic Technology (AST) Program focusing on subsonic test conditions. The test rig is currently configured to test a fuel-rich, catalytic reaction experiment, however, it is soon to be converted over to a flame tube type rig. The test section configuration is 100 in. in length and the interface is a 6 in., 900 LB inlet flange and a 6 in., 150 LB outlet flange. The flame tube rig will have a 3 in. square or circular cross-section flow path which will match the flame tube rig in the test cell CE-5. This new test rig will be used to evaluate and screen fuel injector concepts prior to testing at the higher pressure facilities. The advantages of using this lower pressure facility are its lower cost and its availability. This facility can, therefore, be used to select the more promising concepts to be tested at the higher demand facilities.

The facility is capable of supplying heated air to the test rig at 125 psig (8.5 ATM), 1500 °F and 2.0 pps; and the facility uses a shell and tube heat exchanger to heat the air. The flame tube test rig will be operated at 90 psig. The hot exhaust air within the combustion zone is limited to 3000 °F and is quenched with water spray before venting to the atmosphere. The vitiated air in the heat exchanger, 3.0 pps at 125 psig, is heated through the use of a NASA designed, jet fueled burner can to 1650 °F. Two fuel storage tanks, one 600 gallon tank and one 1000 gallon tank, are located at the facility and a maximum flow of 5.5 GPM at 480 psig can be supplied to the test rig. Cooling water, 118 GPM at 115 psig, is available for cooling and quenching purposes.